Amendments to the Claims:

Applicants reserve the right to pursue any canceled claims at a later date.

The following listing of claims will replace all prior versions, and listings, of claims in the application:

1 – 14 (canceled)

15. (currently amended) A turbine blade, comprising:

a blade leaf arranged along a blade axis having a blade tip, a root opposite the tip, a suction side and a pressure side;

a platform region arranged at the root of the blade leaf; and

a platform arranged at the platform region having a width and extending transversely with respect to the blade axis and partially formed by a first sheet metal component secured to a first abutment arranged on the blade leaf such that the first sheet metal component forms a seal when installed between the first abutment and a second abutment arranged on an axially adjacent turbine blade, wherein the first abutment and the second abutment are each configured as a radial groove protruding in an axial direction of the rotor sufficient to resist an operative radial force of the respective first sheet metal component and the second sheet metal component.

16. (previously presented) The turbine blade as claimed in claim 15, wherein the first sheet metal component is resilient and elastic.

17. (previously presented) The turbine blade as claimed in claim 16, wherein the second abutment is arranged directly on an adjacent turbine blade.

18. (previously presented) The turbine blade as claimed in claim 17, wherein the platform comprises a second sheet metal component secured to a third abutment arranged on a side of the blade leaf opposite that of the first abutment.

19. (previously presented) The turbine blade as claimed in claim 18, wherein the second sheet metal component is formed from a resilient elastic material.

20. (previously presented) The turbine blade as claimed in claim 19, wherein each abutment is a groove or edge.

21. (previously presented) The turbine blade as claimed in claim 15, wherein the second abutment is a bearing support.

22. (previously presented) The turbine blade as claimed in claim 15, wherein the first component is not secured to the second abutment when the turbine is not operational.

23. (previously presented) The turbine blade as claimed in claim 15, wherein during the rotary operation of a rotating turbine blade a self-generated centrifugal force acting radially outward along the blade axis is generated as a result of the blade rotation and the first sheet metal component is pressed against the second abutment by the self-generated force.

24. (previously presented) The turbine blade as claimed in claim 15, wherein the platform region has a blade foot as a load-bearing structure.

25. (currently amended) A gas turbine, comprising:

a flow duct extending along an axis of the turbine having an annular cross section for a working medium; and

a plurality of blade stages having a plurality of annularly arranged turbine blades that extend radially into the flow duct arranged one after another along the axis of the turbine, wherein each turbine blade comprises:

a blade leaf arranged along a blade axis having a blade tip, a foot opposite the tip, a suction side and a pressure side;

a platform region arranged at the foot of the blade leaf; and

a platform arranged at the platform region having a width and extends transversely with respect to the blade axis and partially formed by a first resilient and elastic material secured to a first abutment arranged on the blade leaf such that the first resilient elastic component forms a seal when installed between the first abutment and a second abutment arranged with an axially adjacent turbine blade, wherein the first abutment and the second abutment are each configured as a radial groove protruding in an axial direction of the rotor sufficient to resist an operative radial force of the respective first sheet metal component and the second sheet metal component.

26. (previously presented) The gas turbine as claimed in claim 25, wherein the first resilient elastic material is formed from sheet metal.

27. (previously presented) The gas turbine as claimed in claim 25, wherein, during the rotary operation of a rotating turbine blade a centrifugal force acting from the foot of the blade leaf in the direction of the blade leaf is generated as the result of blade rotation, the first resilient elastic material is pressed against the second abutment by the centrifugal force is fastened to the second abutment by the centrifugal force.

28. (previously presented) The gas turbine as claimed in claim 25, wherein during the operation of a stationary turbine blade in the form of a stationary guide blade, a pressure drop from the foot of the blade leaf acting in the direction of the blade leaf is generated by a cooling medium, the first resilient elastic material is pressed against the second abutment by the pressure drop and is fastened against the second abutment by the resulting pressure.

29. (previously presented) The gas turbine as claimed in claim 25, wherein the first resilient elastic material operatively functions as a sealing element.

30. (previously presented) The gas turbine as claimed in claim 25, wherein a continuous boundary of the flow duct is formed between a first turbine blade and an adjacent second turbine blade of the same blade stage by the first resilient elastic component of the first turbine blade and by a second resilient elastic material of a second turbine blade.

31. (previously presented) The gas turbine as claimed in claim 30, wherein the first and second resilient elastic materials are formed from sheet metal.

32. (previously presented) The gas turbine as claimed in claim 25, wherein a continuous boundary of the flow duct is formed between a first turbine blade of the first blade stage and a second turbine blade of a second blade stage axially adjacent to the first turbine blade, by the first resilient elastic material of the first turbine blade and by a second resilient elastic material of a second turbine blade.

33. (previously presented) The gas turbine as claimed in claim 25, wherein a first resilient elastic material arranged on a first turbine blade and a second resilient elastic material arranged on a second turbine blade are both secured at a further abutment of either the first or second turbine blades.

34. (currently amended) A turbine blade, comprising:

a blade airfoil section arranged along a blade axis having a blade tip, a root opposite the tip, a pressure side and a suction side opposite the pressure side;

a platform region arranged at the root of the blade leaf; and

a platform arranged at the platform region having a width and extending transversely with respect to the blade axis and partially formed by a first bendable metal component secured to a first abutment arranged on the blade leaf such that the sheet metal component forms a seal when installed between the first abutment and a second abutment arranged on an axially adjacent turbine blade, wherein the first abutment and the second abutment are each configured as a radial groove protruding in an axial direction of the rotor sufficient to resist an operative radial force of the respective first sheet metal component and the second sheet metal component.